

# WT-9 Dynamic Speed S

# Weight & Balance

Registration: **PH - 4G3**

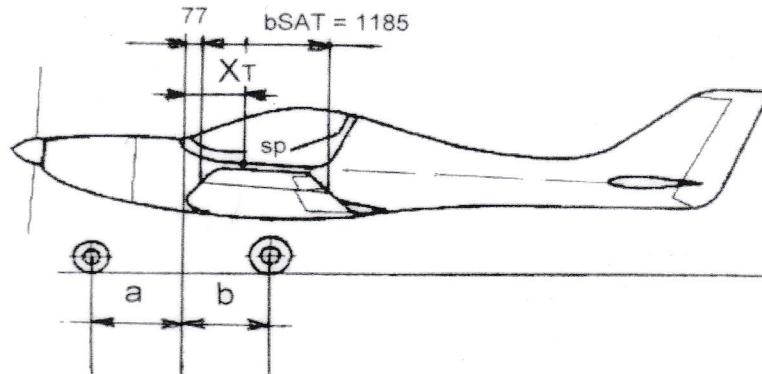
Flight manual Section 6 page 6-3

Datum:		Tijd:	
Vlieger:		License number:	

1e Passagier:	
Adres:	
Woonplaats:	
Telefoon nr:	

**Configuration:** (Empty weight including the operating fluids of the engine and standard equipment).

**Datum point (DP):** leading edge of wing root section



Weighing weights (kg):

Weight and Balance		Weight (kg): instant from DP (mr		Moment (kg mm.)	
Empty aeroplane					
Right wheel	Gr	b=	675		0,0
Left wheel	Gl	b=	675		0,0
Nose wheel	Gn	a=	-748		0,0
<b>Total empty Weight:</b>			<b>322,1</b>	<b>Moment:</b>	<b>63164,3 kgmm</b>
<i>(Oil and coolant including)</i>					
<b>Maximum crew weight 190 kg.</b>					
<b>Maximum weight in baggage compartment 10 kg.</b>					
Crew (two persons)	G crew		720		
Fuel ltr. : <input type="text"/>	0,72 kg/ltr. G Fuel		240		
Bagage:	G bag		1100		
<b>Total weight:</b>		<b>G =</b>	<input type="text"/>	<b>Total Moment:</b> <input type="text"/>	

CENTER OF GRAVITY (M of DATUM POINT) is total moment/ total weight of the airplane

$$X_T = \frac{\text{Total moment}}{\text{Total weight}} = \frac{0,0}{0,0} = \text{mm.}$$

C.G. v % MAC (MAC = 1185 mm.)

$$X_T - 77 = \text{mm.}$$

$$X_{CT} \% \text{ MAC} = \frac{X_T - 77}{\text{SAT}} \times 100 \% = \frac{\text{mm.}}{1185} \times 100 \% = \% \text{ MAC}$$

(SAT = 1185)

Permitted C.G. range of empty aeroplane Xct is 12% +\_ 2 % MAC

**Permitted position of C.G in flight is 20 a 30 % MAC**

Signature