

DYNAMIC WT9

Weight & Balance

Registration: PH - 4B7

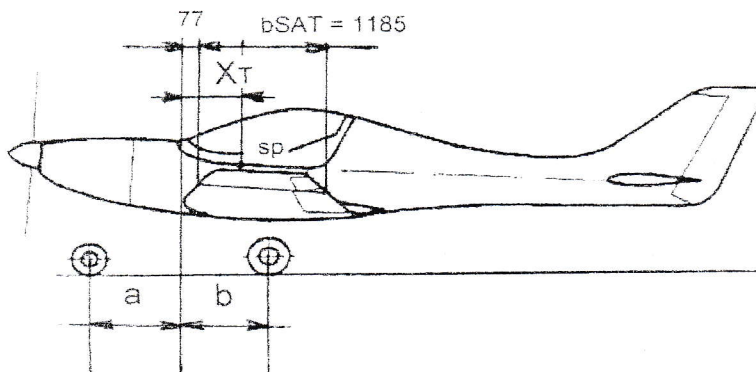
Flight manual Section 6 page 6-3

Datum:		Tijd:	
Vlieger:		License number:	

1e Passagier:	
Adres:	
Woonplaats:	
Telefoon nr:	

Configuration: (Empty weight including the operating fluids of the engine and standard equipment).

Datum point (DP): leading edge of wing root section



Weighing weights (kg):

Weight and Balance

Empty aeroplane	Weight (kg):	Distant from DP (mm)	Moment (kg mm.)
Nose wheel	94,6	a= -722	-68301,2
Right main wheel	103,0	b= 678	69834,0
Left main wheel	104,7	b= 678	70986,6
Total empty Weight:	302,3	Moment:	72519,4

(Oil and coolant including)

Maximum crew weight 190 kg.

Maximum weight in baggage compartment 10 kg.

Crew (two persons)	<input type="text"/>	720	<input type="text"/>
Bagage:	<input type="text"/>	1100	<input type="text"/>
Zero fuel weight:	<input type="text"/>		

Fuel Ltr.	<input type="text"/>	0,72 kg/ltr.	<input type="text"/>	240	<input type="text"/>
Total weight:	G =		<input type="text"/>	Total Moment:	<input type="text"/>

CENTER OF GRAVITY (M of DATUM POINT) is total moment/ total weight of the airplane

$$X_T = \frac{\text{Total moment}}{\text{Total weight}} = \frac{\text{Total moment}}{\text{Total weight}} \text{ mm.}$$

C.G. v % MAC (MAC = 1185 mm.)

$$X_T - 77 = \text{Total moment} - 77 \text{ Total weight} = \text{Total moment} - 77 \text{ Total weight} \text{ mm.}$$

$$X_{CT} \% \text{ MAC} = \frac{X_T - 77}{\text{SAT}} \times 100 \% = \frac{\text{Total moment} - 77 \text{ Total weight}}{1185} \times 100 = \text{Total moment} - 77 \text{ Total weight} \%$$

(SAT = 1185)

Permitted position of C.G in flight is 20 a 30 % MAC